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MANDATORY

**SERVICE
BULLETIN**

MANDATORY BULLETIN Z 142C/6a

1. **AFFECTED AIRCRAFT :** All Z 142C aircraft which should be approved for IFR flights.
2. **SUBJECT :** Provision of the airplane with the Alternate Static Source L 242.8259, which conforms with the applicable requirements of FAR 23.1325.
3. **CORRECTION :**
 - a) Modify the Alternate Static Source of the L 242.8257 type to the L 242.8259 type.
 - b) Replace pages in the Flight Manual and in the Maintenance Manual :
 - A) The Z 142C Flight Manual:
 - a) Remove original pages:
0-5/0-6, 2-25/2-26, 3-1, 3-9/3-10
 - b) Insert new pages:
0-5/0-6, 2-25/2-26, 3-1, 3-9/3-10
 - B) The Z 142C Maintenance Manual, (Vol.I):
 - a) Remove original pages:
0-3/0-4, 2-37/2-38, 2-43/2-44
 - b) Insert new pages:
0-3/0-4, 2-37/2-38, 2-43/2-44
4. **TIME OF COMPLIANCE:** Immediately after the receipt of the Bulletin.
5. **PERFORMED BY :**
 - a) The Manufacturer or any appropriately Authorized Organization shall carry-out the modification.
 - b) The aircraft user shall replace affected pages of Manuals by the new ones.
6. **EXPENSES COVERED BY:**
 - a) The Manufacturer.
 - b) No cost arise.
7. **NECESSARY MATERIAL :**
 - a) The necessary material acc. to the Appendix is enclosed to this Bulletin.
 - b) New pages of the Flight Manual and the Maintenance Manual are enclosed to this Bulletin.
8. **PROCEDURE :**
 - a) The modification procedure is described in the Appendix.
 - b) Replace the original pages by the new ones and enter the revision into the appropriate Record of revisions.

9. THE BULLETIN COMES IN FORCE WITHIN 30 DAYS FROM THE DATE OF APPROVAL BY THE CAI OF THE CZECH REPUBLIC.

10. APPENDICES :
- a) The list of necessary material and the modification procedure (4 pages)
 - b) New pages:
 - A. Z 142C Flight Manual:
0-5/0-6, 2-25/2-26, 3-1, 3-9/3-10
 - B. Z 142C Maintenance Manual, VOL.I:
0-3/0-4, 2-37/2-38, 2-43/2-44.

Antonín Partika
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Manufacturer's
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Milan Vyhnálek
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CAI Representative

28.4.1994
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Date of approval

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APPENDIX No. 1

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1. The Alternate static pressure source-modification from the L 242.8257 type to the L 242.8259 type.

1.1. The material necessary for 1 aircraft:

Item:	Part:	Designation:	Amount:
1	Alternate static pressure source probe	L 242.8258-00.05	1 pc
2	Probe cover	L 242.8258-00.06	1 pc
3	Sealing ring	8x12 ČSN 02 9310.3	1 pc
4	Nut	M8x1 ČSN 02 1403.24	1 pc
5	Valve body	L 242.8259-00.29	1 pc
6	Adjustment screw	L 242.8259-00.30	1 pc
7	Sealing ring	12x16 ČSN 02 9310.3	1 pc
8	Nut	M12x1,5 ONL 3205.2	1 pc
9	Hose	5 PND 21/403-75	L=0,050 m
10	Hose	5 PND 21/403-75	L=0,100 m
11	Hose	5 PND 21/403-75	L=0,900 m
12	Triple-branch piece	T4 MON 3501	1 pc
13	Placard	850-429.2	1 pc
14	Cover	Z 143.8259-00.31	1 pc
15	Placard	850-460.2	1 pc
16	Screw	2,9x9,5 ČSN 02 1235	2 pc
17	Binding wire	Ø 0,8 PN 22-77-76	L=3,000 m
18	Varnish - red	C 2001/8140	m=0,001 kg

1.2. Procedure:

1. Drill the hole $\varnothing 8,5$ in the firewall according to the Fig.1.
2. Drill 3 holes in the L.H. side panel (above the Alternate Source switch) according to the Fig.2.
3. Install the probe assembly (1-4) into the hole $\varnothing 8,5$ on the firewall according to the Fig.3. Bend the probe cover (2) into required shape by bending over the Cylindric body $\varnothing 10$. Lock the nut (4) with a drop of the varnish (18).
4. Mount the Adjusting valve (Items 5-8) into the L.H. side panel above the switch valve according to the Fig.3. Lock the nut (8) with a drop of the varnish (18). Seal the Valve adjustment screw (6) against unintentional misadjustment by application of the appropriate (not hardening) plastic cement (BU Paste, LOCTITE 222 et.c.) on the screw thread before the valve assembly.
5. Install the hoses and the triple-branch piece assembly to the Fig.3. Secure hoses with three coils of the binding wire (17).
6. Replace the original placard located on the LUN 7375 valve on the L.H. instrument panel by the new one (13).
7. Complete the modification by the in-flight adjustment of the Alternate static pressure source at following flight configuration: Take-off weight 1090 kg, ventilation ON, heating OFF, flaps retracted, power max. Continuous, IAS 95 kts.

Adjust the adjustment screw (6) at this configuration until the minimum difference in IAS reading after switching-over from standard to Alternate static source is observed (± 1 kt max.).

8. After adjustment lock the adjustment screw (6) with the colour varnish (18) and install the cover (14) and the placard (15), use screws (16) in accordance with the "View P".

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APPENDIX No. 1

PŘÍLOHA 1

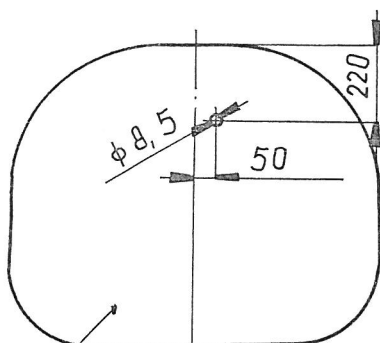
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obr. 1

Fig. 1



Požární stěna

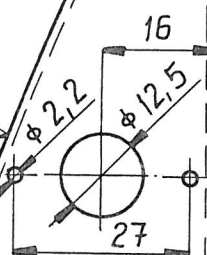
(pohled z motorové strany)

Firewall (view from the engine compartment)

obr. 2

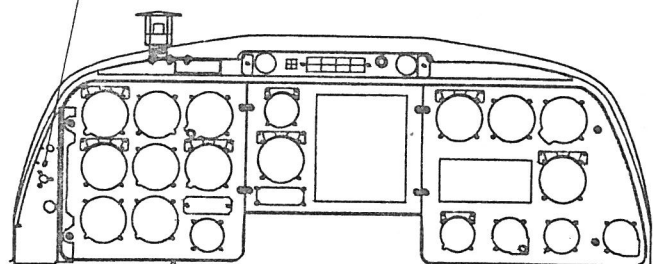
Fig. 2

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Levý boční panel příst. desky

Instrument panel - L.H. side panel



Přístrojová deska

(pohled z pilotního prostoru)

Instrument panel
(view from the pilot's cockpit)

